

Concept: VORTEX

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1. Definition

A vortex is a region in a fluid, where the flow rotates in an organized manner. Such regions are formed naturally when fluid experiences a favorable pressure gradient, i.e., a region of lower pressure is connected to a region where the pressure is higher. Examples abound. Water draining out of a bathtub or washbasin is something we see every day: it naturally develops a swirl as it is "sucked down" through the drain hole. Similar phenomena occur when air near the ground gets heated, and rises: the rising plume does not stay as one parallel set of streamlines, but develops a rotation. Sometimes the rotation can get extremely strong, as in a tornado. On a larger scale, the "eye" of a hurricane clearly shows a vortex. Getting a little more ambitious, we see the same pattern when we look at the arrangement of stars and dust in a galaxy, like our own Milky Way.

2. Introduction

Considering vortices occurring in aircraft applications, such as the tip vortex from a wing or rotor blade, note that the fluid elements inside the vortex do not necessarily remain inside the vortex: there can be continuous exchange of matter through a vortex. Hence the definition of a vortex as a region with certain characteristics, rather than as a distinct entity. Viewed in another sense, the effect of an airfoil on a freestream can be replaced by the effect of a vortex in the freestream: it induces a rotation in the flow.

If something is so prevalent in nature, there must be simple ways of describing the fundamental phenomenon. Consider the following explanation, based on conservation of angular momentum (this follows from Newton's Second Law of Motion): We allow water to collect in a washbasin, and let it sit for some time until it is more-or-less stagnant. Then we open the drain-hole by pushing on the knob which controls it, again a symmetric maneuver: this should affect the fluid all around the drain hole at the same instant, and to the same amount. The water starts flowing into the drain hole (since the pressure is lower at the hole). A "sink" flow develops, one where the flow is moving radially into the hole.

30 Where is the rotation? Well, here is one way by which it can start. Suppose there is a small disturbance in
31 the flow at some radius "r" at time "t", in this sink flow. A "disturbance" in this context is something which
32 makes the flow go a bit sideways, rather than straight along the radius. Now, if one calculates the angular
33 momentum of all the fluid within the radius "r", one finds that most of it has zero angular momentum, but
34 this small element has a tangential "vt" at distance r. So the total angular momentum of the fluid within
35 the radius "r" is "rvt".

36

37 Newton takes over at this point. The angular momentum remains constant unless there is some torque
38 to change it. Away from the solid surface of the washbasin, where there will be some viscous drag, there is
39 no such torque. The vortex is formed. According to our physics so far, the variation of tangential velocity
40 with radial distance is a hyperbolic relationship, as shown below.

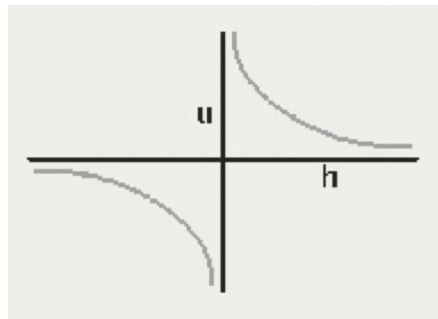


Figure 1: Ideal Vortex Velocity Profile

41 3. Structure

42 What happens at the center? The figure above predicts that tangential velocity should go to +infinity as
43 you approach from one side, and -infinity if you approach from the other side. Obviously, at $r=0$, something
44 drastic must happen because there must be infinite shear strain instead. In the case of a drain-hole, there is
45 no water at the very center: its probably still air. So we don't have to worry whether infinite velocity occurs
46 at zero radius. In the case of a vortex in air, such as a tornado, we do have to worry about this "core" region.
47 Obviously the swirl velocity cannot become infinite at the center of the vortex. Instead, as we get closer
48 to the center, and fluid encounters a sharp change in velocity in a small distance, viscous stress develops.
49 The "velocity discontinuity" gets smoothed out. In this region, the velocity changes smoothly from a high
50 positive at one edge to a high negative at the other, going through zero. This is what happens, for example,
51 if there were a solid cylinder placed in this core region: it would spin, so that its surface velocity matched
52 the tangential velocity of the fluid at its outer radius. The velocity at the center of the spinning cylinder
53 is zero. Thus, in the crudest model of a "two-dimensional vortex", a region of "solid-body rotation" forms

inside this core region. The vortex velocity profile now looks as shown below: A short time later, (time

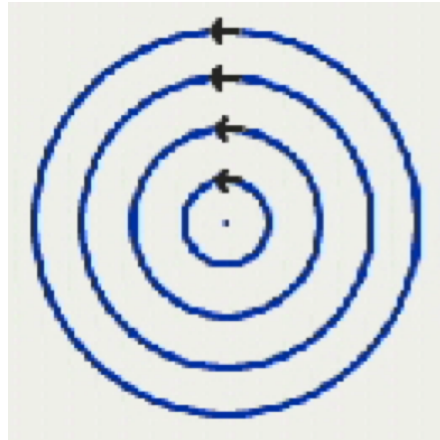


Figure 2: Irrotational Flow Outside the Vortex Core

54

55 $t+dt$), the fluid which was at radius r has moved inward by dr . A circle of radius " $r-dr$ " will enclose the
56 fluid containing the disturbance. Since angular momentum is conserved, the total angular momentum is
57 still " h ", except that it is now within a smaller radius. The tangential velocity of the fluid must now
58 have increased, inversely proportional to r . The fluid swirls around faster and faster. The magnitude of the
59 angular velocity, given by ω , increases, inversely proportional to the square of the radius. Now we must point
60 to a strange aspect. In fluid mechanics, we can go to the ideal 2-D point vortex, and plot streamlines of
61 the fluid. They are all circles (if one ignores the sink, which is a 3-D phenomenon). Thus, we can say that
62 in the ideal 2-D vortex, each fluid element goes traveling in a circle, revolving around the center. Imagine
63 a micro-bus full of tourists going along one of these streamlines. The bus is so narrow that it stays on one
64 streamline. The bus is not crossing the markers of this "streamline" so its not going into a region where the
65 velocity is different (if it did, it would spin). The people sitting on the left side of the bus can see the vortex
66 center always on their left; the people on the right always see the outside regions of the flow. The bus is not
67 actually spinning. Thus, we can say that the flow outside the center of the vortex is "irrotational" . The
68 fluid inside the core is, on the other hand, clearly "rotational". Thus the ideal vortex can be modeled as a
69 region of irrotational fluid, going around a region of rotational fluid, called the core. In ideal fluid dynamics
70 (i.e., "potential flow") we limit our flow domain to that outside the rotational region. Here the "influence"
71 of the vortex in the ideal fluid can be described by the Biot-Savart Law:

72 4. Biot-Savart Law

73 4.1. The 3-D vortex, and a simple view of "vortex breakdown"

74 Now let's return to reality: in the "core" region of a real vortex, there is most certainly a strong "axial"
75 flow: flow along the axis of the vortex. This is the "sink" effect, which produced the vortex in the first place.

76 What causes the sink effect is a matter of circumstance. In the case of a Galactic Black Hole (of which we
77 can speak confidently, since no one who has been there is likely to return and contradict us) gravity provides
78 the "sink" effect: it pulls everything towards it, never to give up anything until a Final Explosion, perhaps.
79 In the case of the kitchen sink, it is again gravity, and the fact that the opening of the drain-hole opened a
80 region of low pressure to which the fluid can flow. What if the drain-hole is stopped suddenly? The pressure
81 gradient becomes less and less favorable, as the liquid level builds up in the drain hole. The axial flow stops.
82 Snap! the rotation also stops. Why? This can be explained by a combination of conservation of mass,
83 and conservation of momentum. When the axial flow stops, the mechanism for strengthening the rotation,
84 described in the equations above, also stops: there is no radial flow of the liquid. In the case of a vortex
85 over a delta wing, a similar phenomenon is observed. The pressure gradient becomes less favorable, and the
86 flow along the axis, which was coming along a thin tubular region which we called the "core", slows down.
87 Conservation of mass dictates that the diameter of the core must increase when the fluid slows down: to
88 accommodate the same mass flow rate, the tube diameter must get bigger. Thus, the edge of the core, where
89 the rotation is fastest, moves out to a larger radius, like an ice dancer stretching her arms. Conservation of
90 angular momentum takes over: the rotation slows down. This is an extremely simplified explanation of the
91 phenomenon of "vortex bursting" or "vortex breakdown", a topic on which debates rage, with the fervor of
92 a religious war. We are not getting into the chicken-or-egg issue of what happens first: does the rotation
93 slow down, forcing an increase in pressure, or vice versa? What was the first disturbance? You can read all
94 the papers on these issues and decide for yourself.

95

96 *4.2. Vortex Interactions*

97 In a real fluid, a vortex might sustain itself for a very long period, because, away from the core, there is
98 very little viscous stress. Thus, the tip vortex left behind by an aircraft might persist in the atmosphere for
99 quite a long time (several minutes). A hurricane might persist for days. When this organized fluid movement
100 encounters other regions (say a solid surface, or another vortex), regions of high shear strain develop, and
101 viscous stresses develop. This might have several kinds of effects.

102

103 *4.3. 3.3 The "Image Vortex" model*

104 We can look at vortex interactions by considering the effect of each vortex at the center of the other.
105 We can consider each vortex to be located in the infinite region of influence of the other vortex, and use the
106 Biot-Savart Law to calculate the "velocity induced at the center of vortex A by the vortex B" for example.
107 This tells us how each vortex moves with respect to the other.

108 A convenient way to look at vortex interactions with a surface is the "image" concept. At a solid surface,
109 there is a no-slip condition. The effect of this is to slow down the vortex flow with respect to the surface;
110 however, this may in fact accelerate the motion of the center of the vortex with respect to the surface. We
111 can model the effect of a solid surface by placing a "mirror image" of the original vortex at a distance from
112 the surface which is correct. For a simple straight wall, the mirror image is just one vortex identical in
113 strength but opposite in sense of rotation to the original vortex, placed at the same distance behind the wall
114 as the original vortex is in front of it. This is shown below:

115

116 **5. Supersets**

117 **6. Subsets**

118 **7. Other fields**

119 **8. Calculators/Applets**

120 **9. 10 Analytical Codes**

121 **10. Notes:**

122 **11. Byline**

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124 **12. References used:**

- 125 1. [1] NASA Thesaurus, Washington, DC: National Aeronautics and Space Administration.
- 126 2. Richards, B.E., Brebner, G.G., Stahl, W., "Missile Aerodynamics". AGARD Lecture Series No. 98,
127 February 1979. Technical Editing and Reproduction Ltd., London.
- 128 3. Brebner, G.G., "A Brief Review of Air Flight Weapons". AGARD Lecture Series No. 98, February
129 1979. Technical Editing and Reproduction Ltd., London, pp. 1-1 - 1-12.
- 130 4. Brebner, G.G., "General Missile Aerodynamics". AGARD Lecture Series No. 98, February 1979.
131 Technical Editing and Reproduction Ltd., London, pp. 2-1 - 2-16.
- 132 5. Stahl, W.H., "Aerodynamics of Low Aspect Ratio Wings". AGARD Lecture Series No. 98, February
133 1979. Technical Editing and Reproduction Ltd., London, pp. 3-1 - 3-64.
- 134 6. Esch, H., "Bodies". AGARD Lecture Series No. 98, February 1979. Technical Editing and Reproduc-
135 tion Ltd., London, pp. 4-1 - 4-29.

- 136 7. Wardlaw, A.B., Jr., "High-Angle-of-Attack Missile Aerodynamics". AGARD Lecture Series No. 98,
137 February 1979. Technical Editing and Reproduction Ltd., London, pp. 5-1 - 5-53.
- 138 8. Delery, J., Sireix, M., "Ecoulements de Culot". AGARD Lecture Series No. 98, February 1979.
139 Technical Editing and Reproduction Ltd., London, pp. 6-1 - 6-78.
- 140 9. Brebner, G.G., "The Control of Guided Weapons". AGARD Lecture Series No. 98, February 1979.
141 Technical Editing and Reproduction Ltd., London, pp. 7-1 - 7-29.
- 142 10. Mathews, C.B., "Store Separation". AGARD Lecture Series No. 98, February 1979. Technical
143 Editing and Reproduction Ltd., London, pp. 8-1 - 8-76.
- 144 11. Richards, B.E., "Kinetic Heating of High Speed Missiles". AGARD Lecture Series No. 98, February
145 1979. Technical Editing and Reproduction Ltd., London, pp. 9-1 - 9-21.
- 146 12. Young, A.D., et al., (Ed): "Aerodynamics of Vortical Type Flows in Three Dimensions". AGARD
147 Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1
- 148 13. Peake, D.J., Tobak, M., "On Issues Concerning Flow Separation and Vortical Flows in Three Dimen-
149 sions". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 1-1 to 1-31.
- 150 14. Hornung, H.G., "The Vortex Skeleton Model for Three-Dimensional Steady Flows". AGARD Con-
151 ference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 2-1 to 2-12.
- 152 15. Roberts, L., "On the Structure of the Turbulent Vortex". AGARD Conference Proceedings No. 342,
153 July 1983. ISBN 92-835-0334-1, pages 3-1 to 3-11.
- 154 16. Strange, C., Harvey, J.K., "Instabilities in Trailing Vortices: Flow Visualization Using Hot-Wire
155 Anemometry". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 4-1 to
156 4-11.
- 157 17. Delery, J., Horowitz, E., "Interaction Entre Une Onde de Choc et Une Structure Tourbillonnaire
158 Entroulee". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 5-1 to 5-19.
- 159 18. Poll, D.I.A., "On the Generation and Subsequent Development of Spiral Vortex Flow Over a Swept-
160 Back Wing". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 6-1 to
161 6-14.
- 162 19. Verhaagen, N.G., "An Experimental Investigation of the Vortex Flow over Delta and Double-Delta
163 Wings at Low Speed". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 7-1
164 to 7-16.
- 165 20. Werle, H., "Visualisation des Ecoulements Tourbillonnaires Tridimensionnels". AGARD Conference
166 Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 8-1 to 8-20.
- 167 21. Vorropoulos, G., Wendt, J.F., "Laser Velocimetry Study of Compressibility Effects on the Flow Field
168 of a Delta Wing". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 9-1 to
169 9-13.

- 170 22. Lamar, J.E., Campbell, J.F., "Recent Studies at NASA-Langley of Vortical Flows Interacting with
171 Neighboring Surfaces". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages
172 10-1 to 10-32.
- 173 23. Erickson, G.E., Gilbert, W.P., "Experimental Investigation of Forebody and Wing Leading Edge
174 Vortex Interactions at High Angles of Attack". AGARD Conference Proceedings No. 342, July 1983. ISBN
175 92-835-0334-1, pages 11-1 to 11-20.
- 176 24. Byram, T., Petersen, A., Kitson, S.T., "Some Results from a Programme of Research into the
177 Structure of Vortex Flow Fields Around Missile Shapes". AGARD Conference Proceedings No. 342, July
178 1983. ISBN 92-835-0334-1, pages 12-1 to 12-20.
- 179 25. Maskew, B., "Predicting Aerodynamic Characteristics of Vortical Flows on Three-Dimensional Con-
180 figurations Using a Surface-Singularity Panel Method". AGARD Conference Proceedings No. 342, July
181 1983. ISBN 92-835-0334-1, pages 13-1 to 12-12.
- 182 26. Vollmers, H., Kreplin, H.P., Meier, H.U., "Separation and Vortical Type Flow Around a Prolate
183 Spheroid - Evaluation of Relevant Parameters". AGARD Conference Proceedings No. 342, July 1983. ISBN
184 92-835-0334-1, pages 14-1 to 14-14.
- 185 27. Evangelou, P., "On the Generation of Vortical Flows At Hypersonic Speeds Over Elliptic Cones".
186 AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 15-1 to 15-9.
- 187 28. Wickens, R.H., "Viscous Three-Dimensional Flow-Separation From High-Wing Propeller Turbine
188 Nacelle Models". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 16-1 to
189 16-29.
- 190 29. Smith, J.H.B. "Theoretical Modeling of Three-Dimensional Vortex Flows in Aerodynamics". AGARD
191 Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 17-1 to 17-21.
- 192 30. Hoeijmakers, H.W.M., "Computational Vortex Flow Aerodynamics". AGARD Conference Proceed-
193 ings No. 342, July 1983. ISBN 92-835-0334-1, pages 18-1 to 18-35. Weiland, C., "Vortex Flow Simulations
194 Past Wings Using the Euler Equations". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-
195 835-0334-1, pages 19-1 to 19-12.
- 196 31. Huberson, S., "Simulation d'Écoulements Turbulents par Une Méthode de Tourbillons Ponctuels".
197 AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 20-1 to 20-8.
- 198 32. Rizzi, A., Eriksson, L-E., Schmidt, W., Hitzel, S., "Numerical Solutions of the Euler Equations
199 Simulating Vortex Flows Around Wings". AGARD Conference Proceedings No. 342, July 1983. ISBN
200 92-835-0334-1, pages 21-1 to 21-14.
- 201 33. Steinhoff, J., Ramachandran, K., Suryanarayanan, K., "The Treatment of Convected Vortices in
202 Compressible Potential Flow". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1,
203 pages 22-1 to 22-12.

- 204 34. Leibovich, S., "Vortex Stability and Breakdown". AGARD Conference Proceedings No. 342, July
205 1983. ISBN 92-835-0334-1, pages 23-1 to 23-22.
- 206 35. Persen, L.N., "The Break-Up Mechanism of a Streamwise Directed Vortex". AGARD Conference
207 Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 24-1 to 24-5.
- 208 36. Escudier, M.P., Keller, J.J., "Vortex Breakdown: A Two-Stage Transition". AGARD Conference
209 Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 25-1 to 25-8.
- 210 37. Krause, E., "A Contribution to the Problem of Vortex Breakdown". AGARD Conference Proceedings
211 No. 342, July 1983. ISBN 92-835-0334-1, pages 26-1 to 26-4.
- 212 38. Nakamura, Y., Leonard, A., Spalart, P.R., "Numerical Simulation of Vortex Breakdown by the Vortex
213 Filament Method". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 27-1
214 to 27-13.
- 215 39. Solignac, J.L., Leuchter, O., "Etudes Experimentales d'Ecoulements Tourbillonnaires Soumis a des
216 Effets de Gradient de Pression Adverse". AGARD Conference Proceedings No. 342, July 1983. ISBN
217 92-835-0334-1, pages 28-1 to 28-25.
- 218 40. Maxworthy. T., Mory, M., Hopfinger, E.J., "Waves on Vortex Cores and their Relation to Vortex
219 Breakdown". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 29-1 to
220 29-13.
- 221 41. Rao, D.M., "Vortical Flow Management for Improved Configuration Aerodynamics - Recent Experi-
222 ences". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 30-1 to 30-14.
- 223 42. Spillman, J.J., Fell, M.J., "The Effects of Wing Tip Devices on the Performance of the BAe Jet-
224 stream". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 31A-1 to 31A-11.
- 225 43. Spillman, J.J., "The Effect of Wing Tip Devices on the Far-Field Wake of a Paris Aircraft". AGARD
226 Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1, pages 31B-1 to 31B-9.
- 227 44. Baron, A., de Ponte, S., "Boundary Layer Segmentation on Sharp Highly-Swept Loading Edges and
228 its Effects on Secondary Vortices". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-
229 1, pages 32-1 to 32-7.
- 230 45. Seginer, A., Salomon, M., "Augmentation of Fighter Aircraft Performance By Spanwise Blowing
231 Over the Wing Leading Edge". AGARD Conference Proceedings No. 342, July 1983. ISBN 92-835-0334-1,
232 pages 33-1 to 33-26.
- 233 46. Young, A.D., (Ed). "Round Table Discussion", AGARD Conference Proceedings No. 342, July 1983.
234 ISBN 92-835-0334-1, pages RTD1-RTD5.
- 235 47. Batchelor, G.K., "An Introduction to Fluid Dynamics". Cambridge University Press, 1967. Batche-
236 lor, G.K., "Axial Flow in Trailing Line Vortices". Journal of Fluid Mechanics, 20, 4, p. 645-658, 1964.
- 237 48. Crow, S.C., "Stability Theory for a Pair of Trailing Vortices". AIAA Journal, 8, 12, 1970.

- 238 49. Donaldson, C. du P., Sullivan, R.D., "Decay of an Isolated Vortex" . In "Aircraft Wake Turbulence
239 and its Detection", Ed. J. Olson, A. Goldberg, M. Rogers, Plenum Press, NY, 1971, p. 389-412.
- 240 50. Fackrell, J.E., "Some Observations of a Trailing Vortex". M.Sc. Thesis, Imperial College, London,
241 1970.
- 242 51. Graham, J.A.H., Newman, B.G., Phillips, W.R., "Turbulent Trailing Vortex with Central Jet or
243 Wake". ICAS Paper 74-40, 1974.
- 244 52. Govindaraju, S.P., Saffman, P.G., "Flow in Turbulent Trailing Vortex". Physics of Fluids, 14,
245 October 1971, pp. 2074-2080.
- 246 53. Harvey, J.K., "Some Observations of the Vortex Breakdown Phenomenon". Journal of Fluid Me-
247 chanics, 14, p. 585-592, 1962.
- 248 54. Lamb, H., "Hydrodynamics". Cambridge University Press, 1932.
- 249 55. Lambourne, N.C., Bryer, D.W., "The Bursting of Leading Edge Vortices - Some Observations and
250 Discussion of the Phenomenon". ARC R&M 3282, 1962.
- 251 56. Owen, P.R., "The Decay of a Turbulent Trailing Vortex". Aero Quarterly, Vol.21, 1970. Rayleigh,
252 Lord, "On The Dynamics of Revolving Fluids". Proceedings of the Royal Society, London, A, 93, pp.
253 148-154, 1917.
- 254 57. Sarpkaya, T., "An Experimental Investigation of the Vortex-Breakdown Phenomenon". US Naval
255 Postgraduate School, NPS-59SL0071A, 1970.
- 256 58. Spreiter, J.R., Sachs, A.M., "The Rolling Up of the Trailing Vortex Sheet and its Effects of the
257 Downwash Behind Wings". Journal of the Aeronautical Sciences, 18, 1, 21-32, 1951.
- 258 59. Squire, H.B., "The Growth of a Vortex in a Turbulent Flow". British ARC 16666 F.M. 2053, 1954.
259 Also, Aeronautical Quarterly, Vol.16, 1965, pp. 302-306.
- 260 60. Taylor, G.I., "Stability of Viscous Liquid Contained Between Two Rotating Cylinders". Philosophical
261 Transactions of the Royal Society, A, 223, pp. 289-343, 1923.
- 262 61. Iverson, J.D., "Correlation of Turbulent Trailing Vortex Decay Data". AIAA Journal of Aircraft,
263 May 1976.
- 264 62. Roberts, L., "Persistence and Decay of Wake Vorticity; AGARD Conference on Flight/Ground
265 Testing Facilities Correlation, 1975.
- 266 63. Hoffman, E.R., Joubert, P.N., "Turbulent Line Vortices", Journal of Fluid Mechanics, Vol. 16, 1963,
267 pp. 395-411.
- 268 64. Corsiglia, V.R., Schwind, R.G., Chigier, N.A., "Rapid Scanning, Three Dimensional Hot Wire
269 Anemometer Surveys of Wing Tip Vortices". AIAA Journal of Aircraft, Vol.10, No.12, December 1972, pp.
270 752-757.
- 271 65. Donaldson, C. DuP., "Calculation of Turbulent Shear Flow, for Atmospheric and Vortex Motions".

- 272 AIAA Journal, Vol.10, 1972, pp. 4-12.
- 273 66. Tung, C., Pucci, C.S., Caradonna, F.X., Morse, H.A., "The Structure of Trailing vortices Generated
274 By Model Rotor Modes". NASA TM 81316.
- 275 67. Kuchemann, D., "Types of Flow on Swept Wings". Journal of the Royal Aeronautical Society, Vol.
276 57, 1953, pp. 683-699.
- 277 68. Kuchemann, D., Weber, J., "Vortex Motions". ZAMM, Vol. 45, 1965, pp. 457-474. Werle, H., "Sur
278 l'eclatement des tourbillons". ONERA N.T., 175 (1971).
- 279 69. Ludwig, H., "Vortex Breakdown". DLR-FB 70-40, 1970. Hall, M.G., "Vortex Breakdown". Annual
280 Review of Fluid Mechanics, Vol. 4, 1972, pp. 195-218.
- 281 70. Hayashi, Y., Nakaya, T., "Flow field in a vortex with breakdown above sharp-edged delta wings".
282 Japanese National Aerospace Laboratory, TR-423, 1975.
- 283 71. Squire, L.C., "Flow Regimes Over Delta Wings at Supersonic and Hypersonic Speeds". Aeronautical
284 Quarterly, Vol. 27, 1976, pp. 1-4.
- 285 72. Szodruch, J., Ganzer, U., "On the lee-side flow over delta wings at high angle of attack". AGARD-
286 CPP-247, 1978, pp. 21.1-21.7.
- 287 73. Kuchemann, D., "On some three-dimensional flow phenomena of the transonic type". Symp.
288 Transsonicum ed. by K. Oswatitsch, Spinger-Verlag, Berlin, 1964, pp. 218-248.
- 289 74. Rainbird, W.J., Crabbe, R.S., Peake, D.J., Meyer, R.F., "Some examples of separation in three-
290 dimensional flows" Canadian Aeronautical and Space Journal, 1966, pp. 409-423.
- 291 75. Schlichting, H., "A Survey of Some Recent Research Investigations on Boundary Layers and Heat
292 Transfer". Journal of Applied Mechanics, Vol. 38, 1971, pp. 289-300.
- 293 76. Hummel, D., "Zur Umströmung scharfkantiger schlanker Deltaflügel bei grossen Anstellwinkeln". Z.
294 Flugwissenschaft, Vol. 15, 1967, pp. 376-385.
- 295 77. Hummel, D., "On the vortex formation over a slender wing at large angles of incidence". AGARD-
296 CPP-247, 1978, pp. 15.1-15.17.
- 297 78. Stahl, W., Hartmann, K., Schneider, W., "Force and Pressure Measurements on a Slender Delta
298 Wing at Transonic Speeds and Varying Reynolds Numbers". AGARD-CP-83-71, 1971, pp. 9.1-9.12.
- 299 79. Smith, J.H.B., Kurn, A.G., "Pressure Measurements on a Slender Rhombic Cone at Incidence at
300 Mach Numbers from 0.4 to 1.1.
- 301 80. Hummel, D., Srinivasan, P.S., "Vortex Breakdown Effects on the Low Speed Aerodynamic Charac-
302 teristics of Slender Delta Wings in Symmetrical Flow". J. Royal Aeronautical Society, Vol. 71, 1967, pp.
303 319-322.
- 304 81. Schneider, W., Stahl, W., Hartmann, K., "Interferenz-Erscheinungen an einer schlanken Flügel-
305 Rumpf-Anordnung in kompressibler Strömung". DFVLR/AVA-Bericht 73 A 06, 1974.

- 306 82. Gersten, K., Hummel, D., "Untersuchungen über den Einfluss der Vorderkantenform auf die aerody-
307 namischen Beiwerte schiebender Pfeil- und Deltaflügel von kleinem Seitenverhältnis." DLR-FB 66-86, 1966.
- 308 83. Henderson, W.P., "Effects of Wing Leading-Edge Radius and Reynolds Number on Longitudinal
309 Aerodynamic Characteristics of Highly Swept Wing-Body Configurations at Subsonic Speeds". NASA TN
310 D-8361, 1976.
- 311 84. Bartlett, G.E., Vidal, R.J., "Experimental Investigation of Influence of Edge Shape on the Aerody-
312 namic Characteristics of Low-Aspect-Ratio Wings at Low Speeds". J. Aeronautical Sciences, Vol. 22, 1955,
313 pp. 517-533.
- 314 85. Gersten, K., Decken, J.V.D., "Aerodynamische Eigenschaften schlanker Flügel in Bodennahe, WGLR
315 Jahrbuch 1966, pp. 108-125.
- 316 86. Decken, J.V.D., "Berechnung der Druckverteilung an schlanken dicken Flugkörpern in Bodennahe".
317 Dissertation, TU Braunschweig, 1969.
- 318 87. Ermolenko, S.D., Ragazin, Yu. A., Rogachev, G.V., "Application of the nonlinearity theory of a
319 lifting surface to the calculation of aerodynamic characteristics of a triangular wing moving close to the
320 earth's surface". AD 785154, FTD-HC-23-1802-74.
- 321 88. Fox, C.H., "Prediction of Lift and Drag for Slender Sharp-Edge Delta Wings in Ground Proximity".
322 NASA TN D-4891, 1969.
- 323 89. Engineering Sciences Data, "Low-Speed Longitudinal Aerodynamic Characteristics of Slender Wings".
324 Item Number 71006.
- 325 90. Engineering Sciences Data, "Low-Speed Normal Force and Pitching Moment of Slender Wings in
326 Ground Effect". Item Number 71007.
- 327 91. Harvey, J.K., "Some Measurements on a Yawed Slender Wing With Leading Edge Separation". ARC
328 RM 3160, 1961.
- 329 92. Hummel, D., Redeker, G., "Über den Einfluss des Aufplatzens der Wirbel auf die aerodynamischen
330 Beiwerte von Deltaflügeln mit kleinem Seitenverhältnis beim Schiebeflug". WGLR Jahrbuch 1967, pp. 232-
331 240.
- 332 93. Schlottmann, F., "Stationaire und instationaire Rollmomentenderivative schlanker Flügel in Rollbe-
333 wegung". Z. Flugwiss., Vol.22, 1974, pp. 331-344.
- 334 94. Parker, A.G., "Measurements on a Delta Wing in Unsteady Flow". Journal of Aircraft, Vol. 14,
335 1977, pp. 547-552.
- 336 95. Cross, E.J., "Experimental and Analytical Investigation of the Expansion Flow Field Over a Delta
337 Wing at Hypersonic Speeds". ARL 68-0027, 1968.
- 338 96. Rao, D.M., Whitehead, A.H.,J., "Lee-Side Vortices on Delta Wings at Hypersonic Speeds". AIAA
339 Journal, Vol. 10, 1972, pp. 1458-1465.

- 340 97. Tosti, L.P., "Low Speed Static Stability and Damping-in-Roll Characteristics of Some Swept and
341 Unswept Low-Aspect-Ratio Wings". NACA TN 1468, 1947. Roy, M., "Caracteres de l'ecoulement autor
342 d'une aile en fleche accentuee", C.R. Acad. Sci., Vol. 234, No. 26, Paris, 1952, pp. 2501-2504.
- 343 98. Ornberg, T., "A Note on the Flow Around Delta Wings". Kungl. Tekn. Hogsk., Stockholm, Aero
344 TN 38, 1954.
- 345 99. Fink, P.T., "Wind Tunnel Tests on a Slender Delta Wing at High Incidence". Z. Flugwiss., Vol. 4,
346 1956, pp. 247-249.
- 347 100. Marsden, D.J., Simpson, R.W., Rainbird, W.J., "The Flow Over Delta Wings at Low Speed With
348 Leading Edge Separation". College of Aeronautics, Cranfield, Rep. 114, 1958.
- 349 101. Emerson, H.F., "Wind-tunnel investigation of the effect of clipping the tips of triangular wings of
350 different thickness, camber, and aspect ratio - Transonic bump method". NACA TN 3671, 1956.
- 351 102. Young, A.D., "Some Special Boundary Layer Problems". Z. Flugwiss. Weltraumforsch., Vol. 1,
352 1977, pp. 401-414.
- 353 103. Winter, H., "Stromungsvorgange an Platten und profilierten Korpern bei kleinen Spannweiten".
354 Forsch. Ing.-Wes., Vol. 6, 1935, pp. 40-50, 67-71. Also, Flow phenomena on plates and airfoils of short
355 span. NACA Rep. 798, 1937.
- 356 104. Scholz, N., "Kraft- und Druckverteilungsmessungen an Tragflachen kleiner Streckung". Forsch.
357 Ing.-Wes., Vol. 16, 1949/50, pp. 85-91, see also, J. Aeronautical Sciences, Vol. 16, 1949, pp. 637-638.
- 358 105. Gersten, K., "Nictlineare Tragflachentheorie fur Rechteckflugel bei inkompressibler Stromung". Z.
359 Flugwiss., Vol. 5, 1957, pp. 276-280.
- 360 106. Schlichting, H., Truckenbrodt, E., "Aerodynamik des Flugzueges". Vol. 2, 2nd Edition, Springer,
361 Berlin-Gottingen-Heidelberg, 1969. Also, Airplane Aerodynamics, McGraw-Hill, Dusseldorf.
- 362 107. Wickens, R.H., "The Vortex Wake and Aerodynamic Load Distribution of Slender Rectangular
363 Plates; The Effects of a 20-degree Bend at Mid-Chord. NRC, Canada, NAE LR-458, 1966.
- 364 108. Ahlborn, F., "Die Wirbelbildung im Widerstandsmechanismus des Wassers, Jb. Schiffbautechn.
365 Ges., Vol. 6, 1905, pp. 67-81.
- 366 109. Ahlborn, F., "Die Widerstandsvorgange im Wasser an Platten und Schiffshorpern. Die Entstehung
367 der Wellen. Jb. Schiffbautechn. Ges., Vol. 10, 1909, pp. 370-431.
- 368 110. Prandtl, L., "Fuhrer durch die Stromungslehre". 6th Ed., Vieweg, Braunschweig, 1965, pp. 326,
369 333-336.
- 370 111. Mabey, D.G., "Beyond the Buffet Boundary". Aeronautical Journal, Vol. 77, 1973, pp. 201-215.
- 371 112. Cornish, J.J., "High Lift Applications of Spanwise Blowing". 7th ICAS Congress, Rome, 1970,
372 ICAS P. 70-09.
- 373 113. Dixon, C.J., "Lift and Control Augmentation by Spanwise Blowing Over Trailing Edge Flaps and

- 374 Control Surfaces". AIAA Paper 72-781, 1972.
- 375 114. Werle, H., Gallon, M., "Controle d'écoulements par jet transversal". Aeron. Astron., No. 34, 1972,
376 pp. 21-33.
- 377 115. Bradley, R.G., Whitten, P.D., Wray, W.O., "Leading-edge-vortex augmentation in compressible
378 flow". Journal of Aircraft, Vol. 13, 1976, pp. 238-242.
- 379 116. Holmboe, V., "The Center of Pressure Position at Low Speed and Small Angles of Attack for Certain
380 Type of Delta Wings". SAAB TN 13, 1953.
- 381 117. Wentz, W.H.J., McMahon, M.C., "Further Experimental Investigation of Delta and Double Delta
382 Wing Flow Fields at Low Speeds". NASA CR 714, 1967.
- 383 118. Krogmann, P., "Experimentelle und theoretische Untersuchungen an Doppeldeltaflugeln". AVA
384 Bericht 68 A 35, 1968.
- 385 119. Hopkins, E.J., Hicks, R.M., Carmichael, R.L., "Aerodynamic Characteristics of Several Cranked
386 Leading Edge WingBody Combinations at Mach Numbers From 0.4 to 2.94. NASA TN D-4211, 1967.
- 387 120. Corsiglia, V.R., Konig, D.G., Morelli, J.P., "Large Scale Tests of An Airplane Model With a Double
388 Delta Wing Including Longitudinal and Lateral Characteristics and Ground Effects". NASA TN D-5102,
389 1969.
- 390 121. Stahl, W., "Zum Einfluss eines Strakes auf das Stromungsfeld eines Deltaflugels ($\alpha=2$) bei schallna-
391 hen Geschwindigkeiten. DLR Mitt. 73-04, 1973, pp. 113-135, also, (in English): ESA TT 175, 1975.
- 392 122. Henderson, W.P., Huffmann, J.K., "Effects of Wing Design on the Longitudinal Aerodynamic
393 Characteristics of a Wing Body Model at Subsonic Speeds". NASA TN D-7009, 1972.
- 394 123. Staudacher, W., "Verbesserung der Manoverleistungen im hohen Unterschall". DLR Mitt. 73-04,
395 1973, pp. 137-158, also, NASA TT-F-15, 406; 1974.
- 396 124. Staudacher, W., Zum Einfluss von Flugelgrundrissmodifikationen auf die aerodynamischen Leistun-
397 gen von Kampflugzeugen. Jahrestagung DGLR/OGFT, Innsbruck, 24-28, 1973, DGLR Nr. 73-71.